

TNI	Kozmická technika Príručka pre analýzu mechanického zaťaženia kozmickej lode	TNI CEN/TR 17603-32-26 31 0540
------------	---	--

Space engineering - Spacecraft mechanical loads analysis handbook

Táto technická normalizačná informácia obsahuje anglickú verziu CEN/TR 17603-32-26:2022.
This Technical standard information includes the English version of CEN/TR 17603-32-26:2022.

Táto technická normalizačná informácia bola oznámená vo Vestníku ÚNMS SR č. 08/22

135501



TECHNICAL REPORT**CEN/TR 17603-32-26****RAPPORT TECHNIQUE****TECHNISCHER BERICHT**

June 2022

ICS 49.035; 49.140

English version

Space engineering - Spacecraft mechanical loads analysis handbook

Ingénierie spatiale - Manuel d'analyse des charges
mécaniques pour vaisseaux spatiauxRaumfahrttechnik - Handbuch zur Analyse von
mechanischen Lasten

This Technical Report was approved by CEN on 13 April 2022. It has been drawn up by the Technical Committee CEN/CLC/JTC 5.

CEN and CENELEC members are the national standards bodies and national electrotechnical committees of Austria, Belgium, Bulgaria, Croatia, Cyprus, Czech Republic, Denmark, Estonia, Finland, France, Germany, Greece, Hungary, Iceland, Ireland, Italy, Latvia, Lithuania, Luxembourg, Malta, Netherlands, Norway, Poland, Portugal, Republic of North Macedonia, Romania, Serbia, Slovakia, Slovenia, Spain, Sweden, Switzerland, Turkey and United Kingdom.



CEN-CENELEC Management Centre:
Rue de la Science 23, B-1040 Brussels

Table of contents

European Foreword.....	10
Preface.....	11
1 Scope.....	13
2 References	14
3 Terms, definitions and abbreviated terms.....	15
3.1 Terms from other documents.....	15
3.2 Terms specific to the present document	16
3.3 Abbreviated terms.....	17
4 Overview of the loads analysis process.....	21
4.1 Introduction.....	21
4.2 Loads cycles.....	22
4.3 Logic and sequence of loads analysis.....	23
4.4 Loads and verification approach (prototype or protoflight).....	24
4.5 Loads and levels of assembly	26
4.6 Mechanical loads for design and verification.....	27
4.6.1 Spacecraft flight environments and dynamic loads.....	27
4.6.2 Vibration environments and frequency range	27
4.6.3 Introduction to analysis and test types for verifying mechanical requirements	28
4.6.4 Static and quasi-static loads.....	30
4.6.5 Static loads test.....	32
4.6.6 Spacecraft-launcher coupled loads analysis	33
4.6.7 Sine vibration	34
4.6.8 Spacecraft design loads and test predictions versus LV/SC CLA results	36
4.6.9 Random vibration and vibro-acoustic environment.....	37
4.6.10 Shock testing	39
4.7 Basic principles, criteria and assumptions in structure and loads verification	40
4.7.1 Introduction	40
4.7.2 Equivalence criteria for loads and environments	40
4.7.3 Criteria for assessing verification loads	42

CEN/TR 17603-32-26:2022 (E)

4.7.4	Main inconsistencies of the loads verification process.....	42
4.8	Notching in sine and random vibration testing.....	43
4.8.1	Introduction	43
4.8.2	Example of requirements	44
4.8.3	Basic principles	44
4.8.4	Response and force limiting	45
4.8.5	Criteria for notching justification	46
4.8.6	Conclusions on notching in sine and random vibration testing	48
4.9	References	49
5	Background on structural dynamics	50
5.1	Introduction.....	50
5.1.1	The dynamic environment.....	50
5.1.2	Types of structural analysis.....	51
5.1.3	List of topics.....	51
5.1.4	Principal notations.....	52
5.2	Dynamic environments – analysis and specifications.....	54
5.2.1	Generalities.....	54
5.2.2	Example - the maiden flight of Ariane 1.....	55
5.2.3	Sine environment	57
5.2.4	Transient environment	62
5.2.5	Random environment.....	65
5.2.6	Sine-equivalent dynamics	74
5.2.7	Combined environments	83
5.3	Dynamic analysis.....	86
5.3.1	Frequency domain analysis.....	86
5.3.2	Modal approach	88
5.3.3	Effective mass models	92
5.3.4	Craig-Bampton models	93
5.4	Coupled analysis and notching in sine tests.....	107
5.4.1	FRF coupling.....	107
5.4.2	Modal approach	108
5.4.3	Simple example	109
5.4.4	Use of the shock response spectrum	111
5.5	Primary and secondary notching.....	115
5.5.1	Modes concerned by primary notching.....	115
5.5.2	Secondary notching	115
5.5.3	Simple example	116
5.5.4	Conclusions on notching in sine tests	118
5.6	Random tests	118

CEN/TR 17603-32-26:2022 (E)

5.6.1	Issues on random tests	118
5.6.2	Mechanical equivalence example.....	119
5.6.3	Notching in random vibration tests	121
5.7	Practical aspects of modal effective masses.....	124
5.8	Conclusions	126
5.9	References	126
6	Launcher / spacecraft coupled loads analysis	129
6.1	Introduction.....	129
6.1.1	General aspects.....	129
6.1.2	Launch loads and terminology used in the CLA process	130
6.1.3	The role of the CLA within the loads cycle.....	132
6.2	The phases of the CLA process.....	133
6.2.1	Introduction	133
6.2.2	Parameters driving the CLA process.....	134
6.2.3	Mathematical model verification and database integration	134
6.2.4	Finite element model reduction	134
6.2.5	Checks on the Craig-Bampton matrices and OTM	135
6.2.6	Frequency cut-off for computed modes	135
6.2.7	Coupling of the launcher and spacecraft models.....	135
6.2.8	Calculation of the generalized responses.....	135
6.2.9	Determination of the physical responses.....	135
6.2.10	Post-processing	135
6.2.11	Uncertainty factors	136
6.3	CLA output and results evaluation	137
6.3.1	Overview.....	137
6.3.2	Guidelines to response parameter selection.....	138
6.3.3	Equivalent sine input.....	138
6.3.4	Computation of static components from OTM	138
6.3.5	Relative displacements	139
6.3.6	Interface mechanical fluxes and overfluxes	139
6.3.7	Results review, verification and validation	144
6.3.8	Use of CLA results for structural verification.....	145
6.3.9	Reporting	145
6.4	Ariane 5 coupled loads analysis.....	148
6.4.1	Introduction to Ariane 5 CLA	148
6.4.2	Mission analysis organization and management	149
6.4.3	CLA events and load cases.....	150
6.4.4	Concomitant events and load cases combination.....	159
6.4.5	Flight phases and CLA standard load cases	160

CEN/TR 17603-32-26:2022 (E)

6.4.6	Aspects of the Ariane 5 CLA methodology	163
6.5	The Arianespace spacecraft qualification process	165
6.5.1	Introduction	165
6.5.2	Quasi-static loads	166
6.5.3	Dynamic environment	168
6.6	Space Shuttle coupled loads analysis.....	174
6.6.1	Overview	174
6.6.2	CLA load events.....	175
6.6.3	Elements of the design and verification process for Space Shuttle payloads	176
6.7	References	179
7	Static loads	180
7.1	Introduction.....	180
7.2	Quasi-static loads	180
7.2.1	General aspects.....	180
7.2.2	Equivalence between dynamic conditions and CoG net accelerations	181
7.2.3	Quasi-static loads specification	182
7.2.4	Prediction of QSL and mechanical environment by base-drive analysis.....	184
7.3	Static test philosophy and objectives	184
7.4	Definition of static test configuration and load cases	185
7.4.1	Introduction	185
7.4.2	Boundary conditions.....	186
7.4.3	Loading systems	186
7.4.4	Load cases	187
7.4.5	Instrumentation	188
7.5	Static test evaluation.....	188
7.6	References	190
8	Sine vibration.....	199
8.1	Introduction.....	199
8.2	Sine vibration levels specification	199
8.2.1	Sine loads for spacecraft.....	199
8.2.2	Sine loads for payload and equipment	200
8.3	Simulation / test prediction.....	201
8.3.1	Introduction	201
8.3.2	Boundary conditions.....	201
8.3.3	Damping	202
8.3.4	Notch assessment.....	202
8.4	Sine vibration test	203

CEN/TR 17603-32-26:2022 (E)

8.4.1	Objectives	203
8.4.2	Notching process	204
8.4.3	Test preparation	206
8.4.4	Sine test campaign.....	219
8.5	References	229
9	Random vibration and vibro-acoustics	230
9.1	Introduction.....	230
9.1.1	Overview	230
9.1.2	Random vibration loads	231
9.1.3	Vibro-acoustic loads.....	231
9.2	Requirements	234
9.3	Random vibration specification	234
9.3.1	Introduction	234
9.3.2	Component vibration environment predictor, Spann method	234
9.3.3	Specifications derived from random and vibro-acoustic test data	236
9.3.4	VibroSpec	238
9.3.5	Test/analysis extrapolation method	240
9.4	Random vibration analysis.....	242
9.4.1	Finite element analysis and Miles' equation	242
9.4.2	Finite element analysis.....	243
9.4.3	Guidelines for FE random vibration response analysis	245
9.5	Random vibration testing	247
9.5.1	Introduction	247
9.5.2	Notching.....	247
9.6	Vibro-acoustic analysis	264
9.6.1	Introduction	264
9.6.2	Boundary element analysis	264
9.6.3	Statistical energy analysis.....	267
9.6.4	General guidelines for vibro-acoustic analyses	270
9.7	Acoustic testing	272
9.7.1	Introduction	272
9.7.2	Test plan/procedure	272
9.8	Verification of compliance	274
9.8.1	General aspects.....	274
9.8.2	An example based on the vibration response spectrum	275
9.9	Special topics in random vibration	278
9.9.1	Simulation of the random time series	278
9.9.2	Prediction of random acoustic vibration of equipment mounted on panels	281

CEN/TR 17603-32-26:2022 (E)

9.9.3 Quick way to predict fatigue life (Steinberg method).....	285
9.10 References	288
10 Shock.....	291
10.1 Introduction.....	291
10.2 Shock environment.....	291
10.3 Shock design and verification process	292
10.3.1 Shock input derivation to subsystems	293
10.3.2 Shock verification approach	294
10.3.3 Shock damage risk assessment.....	297
10.4 References	300
11 Dimensional stability.....	301
11.1 Introduction.....	301
11.2 Dimensional stability analysis	302
11.2.1 Thermo-elastic distortion analysis	303
11.2.2 1g-0g transition (gravity release).....	310
11.2.3 Moisture absorption / release	311
11.3 Dimensional stability verification	313
11.3.1 Introduction	313
11.3.2 Thermal distortion test.....	313
11.3.3 Gravity release test	319
11.4 Material property characterisation testing	320
11.4.1 Coefficient of Thermal Expansion (CTE) characterisation	320
11.4.2 Coefficient of Moisture Expansion (CME) characterisation	321
11.5 References	322
12 Fatigue and fracture control.....	323
12.1 Introduction.....	323
12.2 Definitions.....	326
12.3 List of events	326
12.4 Load spectra per event	330
12.4.1 General.....	330
12.4.2 Existing load curves	330
12.4.3 Measured load curves.....	332
12.4.4 Calculating load curves	334
12.5 Generation of fatigue spectra.....	337
12.6 References	339
13 Micro-gravity and micro-vibrations.....	341
13.1 Introduction.....	341

CEN/TR 17603-32-26:2022 (E)

13.1.1	Background.....	341
13.1.2	Scope.....	342
13.2	Micro-gravity	342
13.2.1	General aspects.....	343
13.2.2	Derivation of micro-gravity specifications	354
13.2.3	Micro-gravity environment verification	362
13.3	Micro-vibration	369
13.3.1	General aspects.....	369
13.3.2	Micro-vibration analysis.....	371
13.3.3	Micro-vibration budget assessment.....	379
13.3.4	Pointing error synthesis.....	382
13.3.5	Micro-vibration verification test.....	383
13.4	Micro-gravity and micro-vibration disturbance sources	387
13.4.1	Scope.....	387
13.4.2	Review of potential disturbance sources	387
13.4.3	Characterisation of the disturbance sources forcing functions.....	398
13.5	References	435
14	Soft stowed packaging.....	437
14.1	Introduction.....	437
14.2	Packaging guidelines.....	438
14.3	Materials for packaging.....	439
14.3.1	Physical properties.....	439
14.3.2	Attenuation data for foam packed items	445
14.4	Soft stowed equipment verification flow	451
14.4.1	Hardware categories and criticality.....	451
14.4.2	General verification aspects	451
14.4.3	Off-the-shelf (OTS) items and already existing equipment	456
14.4.4	New equipment / hardware	458
14.2	References	463
15	Nonlinear structures.....	464
15.1	Introduction.....	464
15.2	Common spacecraft structure nonlinearities	464
15.2.2	Damping	465
15.2.3	Contact	466
15.2.4	Nonlinear stiffness.....	467
15.3	Nonlinearity detection	468
15.4	Handling of spacecraft structure nonlinearities.....	469
15.4.1	Introduction	469

CEN/TR 17603-32-26:2022 (E)

15.4.2	Guidelines for testing	470
15.4.3	Nonlinearity characterisation and parameter estimation	472
15.4.4	Guidelines for structure modelling and analysis	474
15.4.5	Impact of nonlinearities on CLA flight load predictions	479
15.5	References	481
16	Finite element models	482
16.1	Introduction	482
16.2	Requirements for structure mathematical models	483
16.3	Introduction to V&V in computational mechanics	483
16.4	Spacecraft finite element model complexity and validation test	486
16.5	Uncertainty quantification during load cycles	487
16.5.1	Overview	487
16.5.2	Dynamic variability or uncertainty factor K_v	487
16.5.3	Model factor K_M	489
16.6	Verification and quality assurance for spacecraft finite element analysis	489
16.7	Mathematical model validation	491
16.7.1	General concepts and terminology	491
16.7.2	Why a mathematical model validation process	492
16.7.3	Categorization of the uncertainty and sources of disagreement between simulation and experimental outcomes	493
16.7.4	Specific aspects of the validation of spacecraft FEM for coupled loads analysis	493
16.7.5	Error localization and model updating by sensitivity and optimization	498
16.7.6	Specific aspects concerning base-drive sine vibration testing and “real-time” model validation	499
16.7.7	Stochastic approaches for model validation	500
16.8	References	501

European Foreword

This document (CEN/TR 17603-32-26:2022) has been prepared by Technical Committee CEN/CLC/JTC 5 “Space”, the secretariat of which is held by DIN.

It is highlighted that this technical report does not contain any requirement but only collection of data or descriptions and guidelines about how to organize and perform the work in support of EN 16603-32.

This Technical report (CEN/TR 17603-32-26:2022) originates from ECSS-E-HB-32-26A.

Attention is drawn to the possibility that some of the elements of this document may be the subject of patent rights. CEN [and/or CENELEC] shall not be held responsible for identifying any or all such patent rights.

This document has been prepared under a mandate given to CEN by the European Commission and the European Free Trade Association.

This document has been developed to cover specifically space systems and has therefore precedence over any TR covering the same scope but with a wider domain of applicability (e.g.: aerospace).

Preface

The “Spacecraft Mechanical Loads Analysis Handbook” has been developed with the aim to harmonize methodologies, procedures and practices currently applied for the conduct of spacecraft and payloads loads analysis. It makes available to the European Space Community a set of well proven methods, procedures and guidelines for the prediction and assessment of structural design loads and for the evaluation of the test loads. In particular, recent advances in the area of structural dynamics and vibrations, in both methodology and performance, have the potential to make spacecraft system analysis and testing more effective from technical, cost, and hardware safety points of view. However, application of advanced analysis methods varies among the Space Agencies and their contractors. Identification and refinement of the best of these methodologies and implementation approaches has been an objective of the Working Group.

The handbook is intended to be a practical guide rather than a theoretical treatise. The emphasis is on dynamic environments of spacecraft, however other mechanical environments are addressed and often the principles are broad enough to be applicable in many cases to launch vehicles as well. It is assumed that the reader has a general knowledge of spacecraft structures and structural dynamics without necessarily being an expert in these disciplines.

This first edition represents a collection of contributions by a number of engineers from throughout the European Space Community. It reflects the insight gained from their practical experience. The contributions have been harmonised and the handbook completed by the “harmonization team”. The level of treatment varies among topics, depending on the issues each author feels is critical and the overall assessment performed by the harmonization team concerning the level of detail in each topic that is important to the loads analysis process.

The book is not intended as a selfstanding textbook since in some cases it is rather complementary to other ECSS documents and more in general to textbooks and publications on spacecraft structures and structural dynamics. It can be a key tool for spacecraft designers, system and structural engineers who need to find out more about mechanical loads analysis and for those in charge of developing requirements and specifications.

The reader benefits best by reading the book sequentially, although most of the chapters are selfcontained, with references to other parts of the book provided as needed. An overview of the chapters is presented below:

- Chapter 4 gives an overview of the loads analysis process aimed at establishing appropriate loads for design and testing.
- Chapter 5 presents a summary of the principles of structural dynamics addressed throughout the different chapters of the present handbook.
- Chapter 6 addresses the launcher / spacecraft coupled loads analyses performed to check that a spacecraft design is compliant with the overall mechanical environment generated by a launcher during all flight phases and to ensure that the mission can be achieved.

CEN/TR 17603-32-26:2022 (E)

- Chapters 7 to 10 deal with analysis and testing related to the various types of mechanical environments generated by the launcher: static in chapter 7, sine vibrations in chapter 8, random vibrations (including vibro-acoustic environment) in Chapter 9, shocks in chapter 10.
- Chapter 11 is devoted to dimensional stability i.e. the behaviour of highly accurate structures to maintain their dimensions under all kinds of conditions.
- Chapter 12 deals with fracture control and fatigue life verification, discussing the various aspects involved in deriving fatigue load spectra to perform analyses or tests.
- Chapter 13 addresses the micro-gravity and micro-vibration environment for which the spacecraft systems should be designed and operated such that limit acceleration levels are not exceeded.
- Chapter 14 is related to soft stowed equipment and the verification process of items packed in foam, to assess the compatibility of the cargo item with the attenuated environments.
- Chapter 15 tackles the problems generated by a nonlinear behaviour of the structures, which can significantly affect the verification process.
- Chapter 16 addresses the mathematical models used for loads analysis, with emphasis on finite element analysis quality and acceptance of the results.

Funding and resources for the handbook were provided by the European Cooperation for Space Standardization leading to the creation of the initial Working Group. However the number of contributors soon increased with time and substantial additional volunteer support was provided by individuals and organizations.

These additional resources have been crucial to the successful accomplishment of the “handbook project”. All the volunteer contributors that have sacrificed their time are gratefully acknowledged, as well as the contributors that made an effort beyond the allocated resources. This first edition of handbook is thus the result of two and a half years of effort by the “enlarged” Loads Analysis Working Group.

A substantial effort has been made to eliminate mathematical and factual errors. Nevertheless it is possible (and likely) that some errors will be found through readers’ use of the handbook. Detected errors along with any omissions, corrections or comments may be sent to either the ECSS Secretariat or to the addresses below. If, as hoped, the book is of use to the space community, it could be updated and made more useful and practical.

1

Scope

This document recommends engineering practices for European programs and projects. It may be cited in contracts and program documents as a reference for guidance to meet specific program/project needs and constraints.

The target users of this handbook are engineers involved in design, analysis and verification of spacecraft and payloads in relation to general structural loads analysis issues. The current know-how is documented in this handbook in order to make this expertise available to all European developers of space systems.

It is a guidelines document; therefore it includes advisory information rather than requirements.

2 References

Due to the structure of the document, each chapter includes at its end the references called in it.

koniec náhľadu – text ďalej pokračuje v platenej verzii STN